

Bombardier Service Bulletin 8–52–58, Revision A, dated November 17, 2006; and Bombardier Service Bulletin 8–52–61, dated October 20, 2006; for related information.

Material Incorporated by Reference

(n) You must use the following service information to do the applicable actions required by this AD, unless the AD specifies otherwise. The Director of the Federal Register approved the incorporation by reference (IBR) under 5 U.S.C. 552(a) and 1 CFR Part 51 of the following service information on the date specified:

(1) Bombardier Service Bulletin 8–52–58, Revision A, dated November 17, 2006, approved for IBR November 16, 2011;

(2) Bombardier Service Bulletin 8–52–61, dated October 20, 2006, approved for IBR November 16, 2011;

(3) Bombardier Service Bulletin 8–52–54, Revision A, dated November 5, 2004, approved for IBR July 18, 2006 (71 FR 34006, June 13, 2006).

(4) For service information identified in this AD, contact Bombardier Inc., Q-Series Technical Help Desk, 123 Garratt Boulevard, Toronto, Ontario M3K 1Y5, Canada; telephone 416–375–4000; fax 416–375–4539; e-mail thd.qseries@aero.bombardier.com; Internet <http://www.bombardier.com>.

(5) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425–227–1221.

(6) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Renton, Washington, on September 23, 2011.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2011–25770 Filed 10–11–11; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2010–0033; Directorate Identifier 2009–NM–099–AD; Amendment 39–16737; AD 2011–14–02]

RIN 2120–AA64

Airworthiness Directives; The Boeing Company Model 767 Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: The FAA is superseding an existing airworthiness directive (AD) that applies to all Model 767 airplanes. The existing AD currently requires repetitive detailed and high frequency eddy current (HFEC) inspections of the station (STA) 1809.5 bulkhead for cracking, and corrective actions if necessary. This AD expands the inspection area to include the vertical inner chord at STA 1809.5. This AD results from reported fatigue cracking in the vertical inner chord and the forward outer chord while doing the detailed inspection of the horizontal inner chord at STA 1809.5. We are issuing this AD to detect and correct fatigue cracking in the bulkhead structure at STA 1809.5 and the vertical inner chord at STA 1809.5, which could result in failure of the bulkhead structure for carrying the flight loads of the horizontal stabilizer, and consequent loss of controllability of the airplane.

DATES: This AD becomes effective November 16, 2011.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in the AD as of November 16, 2011.

ADDRESSES: For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H–65, Seattle, Washington 98124–2207; *phone:* 206–544–5000, extension 1; *fax:* 206–766–5680; *e-mail:* me.boecom@boeing.com; *Internet:* <https://www.myboeingfleet.com>.

Examining the AD Docket

You may examine the AD docket on the Internet at <http://www.regulations.gov>; or in person at the Docket Management Facility between 9 a.m. and 5 p.m., Monday through Friday, except Federal holidays. The AD docket contains this AD, the regulatory evaluation, any comments received, and other information. The address for the Docket Office (telephone 800–647–5527) is the Document Management Facility, U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC 20590.

FOR FURTHER INFORMATION CONTACT:

Berhane Alazar, Airframe Branch, ANM–120S, FAA, Seattle Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; *phone:* 425–917–6577; *fax:* 425–917–6590; *e-mail:* Berhane.Alazar@faa.gov.

SUPPLEMENTARY INFORMATION:

Discussion

The FAA issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that supersedes AD 2006–24–04, Amendment 39–14833 (71 FR 68432, November 27, 2006). The existing AD applies to all Model 767 airplanes. That NPRM was published in the **Federal Register** on February 8, 2010 (75 FR 6154). That NPRM proposed to continue to require repetitive detailed and HFEC inspections of the STA 1809.5 bulkhead for cracking, and corrective actions if necessary. That NPRM also proposed to expand the inspection area to include the vertical inner chord at STA 1809.5.

Comments

We provided the public the opportunity to participate in the development of this AD. We have considered the comments that have been received on the NPRM (75 FR 6154, February 8, 2010).

Support for the NPRM

Continental Airlines (CAL) stated that it supports the intent of the NPRM (75 FR 6154, February 8, 2010).

Request To Revise Paragraph (k)(1) of the NPRM

Boeing requested that we revise the compliance time in paragraph (k)(1) of the NPRM (75 FR 6154, February 8, 2010) to state “whichever occurs later” rather than “whichever occurs first.” Boeing stated that a similar AD, AD 2006–24–04 (71 FR 68432, November 27, 2006) (the AD being superseded), provides a choice of the later of two compliance times. Boeing stated that changing the compliance time language in paragraph (k)(1) of the NPRM would make this AD consistent with AD 2006–24–04.

We agree with the request for the reasons provided by the commenter, and we have revised paragraph (k)(1) of this final rule accordingly.

Request To Add Model 767–300BCF and 767–200SF Structural Repair Manuals (SRMs) To Clarify Terminating Action

Boeing requested that we specify Model 767–300BCF and Model 767–200SF SRMs in paragraphs (i) and (m) of the NPRM (75 FR 6154, February 8, 2010) to clarify the terminating action for converted Model 767–200 and –300 series airplanes. Boeing stated that some Model 767–300 airplanes have been converted to Model 767–300BCF airplanes, and some Model 767–200 airplanes have been converted to Model 767–200SF airplanes. Boeing stated that the Model 767–200SF and Model 767–

300BCF SRMs differ from the Model 767–200 and Model 767–300 SRMs.

We partially agree with the request. The Model 767–300BCF is unique and has its own SRM documents. We have added the Model 767–300BCF SRMs to paragraphs (i) and (j) of this final rule and table 1 of this final rule (table 1 follows paragraph (l) of this final rule; paragraph (l) of this final rule was referred to as paragraph (m) in the NPRM (75 FR 6154, February 8, 2010)).

However, the Boeing Model 767–200SF SRM does not include Subject 53–80–08, which is the subject referenced in paragraphs (i) and (l) of this AD; therefore, we have not changed the AD in regard to this model. The Model 767–200SF is unique and has its own SRM documents. After that material is developed, we will consider requests for approval of an alternative method of compliance (AMOC) under the provisions of paragraph (n) of this final rule.

Request To Clarify and Revise Paragraph (h) of the NPRM

CAL requested that we clarify the AMOC requirements of paragraph (h) of the NPRM (75 FR 6154, February 8, 2010). CAL asserted that paragraph 3.B. of the Accomplishment Instructions of Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009, allows repair of the vertical inner chord in accordance with Repair 11 of Subject 53–80–08; therefore, the vertical inner chord needs to be added to paragraph (h) of the NPRM as an exclusion to the AMOC requirements.

We agree with commenter's request for the reasons provided. We have revised paragraph (h) of this AD to add the reference to the vertical inner chord, as requested by CAL.

CAL also stated that for repairs or replacements of stringers and non-principal structural elements (PSE) parts (e.g., attach brackets, support clips, etc.), an AMOC should not be necessary. For instance, Parts 3 and 4 of Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009, specify the detailed inspection of the structure surrounding the forward outer chord of the STA 1809.5 bulkhead, and the corresponding figures illustrate the inspection area, which encloses the stringers and attach fittings. As written, CAL stated that paragraph (h) of the NPRM (75 FR 6154, February 8, 2010) will prevent operators from making stringer repairs/replacements using “SRM 53–00–03” without first obtaining an AMOC. CAL stated further that replacing cracked clips and brackets using certain “Boeing drawings” should not require an AMOC.

CAL also requested that we revise paragraph (h) of the NPRM (75 FR 6154, February 8, 2010) to read as follows: “If any cracking is found in the skin or the STA 1809.5 bulkhead's principal structural elements (PSE) other than the forward outer chord, horizontal inner chord, and vertical inner chord during any inspection required by paragraph (g) or (k) of this AD, and Boeing Service Bulletin 767–53A0131, dated March 30, 2006; or Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009; specifies to contact Boeing for appropriate action: Before further flight, repair the cracking using a method approved in accordance with the procedures specified in paragraph (n) of this AD. When replacing cracked parts per the Boeing drawings or repairing stringers per 767 SRM 53–00–03, approval in accordance with the procedures specified in paragraph (n) of this AD is not required.”

We partially agree with the request to revise paragraph (h) of this final rule. In order for a method of compliance other than the method(s) provided by the AD to be used, that method must be approved under the provisions of paragraph (n) of this AD. It is crucial that the FAA and Boeing are aware of all repairs made to PSEs or surrounding structure, and that damage tolerance be performed on each repair to establish its effect on the fatigue life of the affected structure.

In addition, we have determined that repairing/replacing the stringers in accordance with the SRM referenced by the commenter is an acceptable method of compliance for those specific requirements of paragraph (h) of this AD. We have revised paragraph (h) of this AD accordingly.

However, we disagree with including references to “Boeing drawings” for replacing cracked clips and brackets. We must cite specific service information (with dates and revision levels) in our ADs and would need to have the applicable drawings submitted for review prior to including those drawings in an AD. Each operator may be using different Boeing drawings as reference for replacing cracked clips and brackets. In addition, we do not consider it appropriate to include various provisions in an AD applicable only to certain airplanes or to a single operator's unique use of an affected airplane. Individual operators may request approval of an AMOC, provided sufficient data are submitted to substantiate such a request.

Request To Clarify the Requirements of Paragraph (l) of the NPRM

ABX requested clarification as to why paragraph (l) of the NPRM (75 FR 6154, February 8, 2010) is included. ABX stated that the NPRM specifies the limits in each applicable paragraph, and that the NPRM does not refer to paragraph 1.E., “Compliance,” of Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009. ABX stated that the inclusion of paragraph (l) of the NPRM leads one to believe that somewhere in the NPRM other limits are hidden.

We agree to provide clarification. We have determined that paragraph (l) of the NPRM (75 FR 6154, February 8, 2010) was included unnecessarily because the NPRM did not refer to paragraph 1.E., “Compliance,” of Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009. Therefore, we have removed that paragraph and other references to it from this final rule.

Request To Revise Note 1 of the NPRM

ABX requested that we revise Note 1 of the NPRM (75 FR 6154, February 8, 2010) from “Guidance on modifying a vertical inner chord * * *” to “Approved methods on modifying a vertical inner chord * * *” ABX stated that Table 1 of the NPRM lists service information that provides guidance on modifying a vertical inner chord. ABX noted that this service information does not appear to be approved by the Manager of the Seattle Aircraft Certification Office (ACO). ABX stated that it believes that this service information should be approved for terminating action for the condition of no cracking found during the most recent detailed and HFEC inspections. ABX noted that the modification, with the removal of the damaged area, is approved as a terminating action when the cracks are found. ABX stated that it believes that the modification should also be approved when no cracks are found.

We partially agree. We agree that accomplishing the terminating modification specified in Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009, terminates the applicable inspections for airplanes on which cracking is found and those on which cracking is not found. We have removed Note 1 of the NPRM (75 FR 6154, February 8, 2010) from the final rule and, instead, have revised paragraph (l) of the final rule to incorporate the information in Note 1 to specify that the optional terminating modification may also be done in

accordance with the applicable Boeing 767 SRM. In addition, paragraph (m) of this final rule specifies that if any cracking is found during any modification done in accordance with paragraph (l) of this AD, and the applicable Boeing 767 SRM specifies to contact Boeing for appropriate action, the cracking must be repaired in accordance with a method approved by the Manager of the Seattle Aircraft Certification Office. We also note that the SRM is an FAA-approved document.

Explanation of Additional Changes to This AD

We have added a new table 1 to this final rule to provide the applicable, current SRMs; we have re-identified subsequent tables accordingly.

We have changed paragraphs (i) and (l) of this final rule to specify specific

steps of the applicable SRM identified in table 1 of this AD. We have also reformatted paragraphs (i) and (l) of this final rule to differentiate the methods of compliance, and added new Note 1 and Note 2 to this final rule to explain the reformatting changes.

We have revised the references to the SRM in the following locations to accurately identify Subject 53–80–08, Fuselage Bulkheads—Section 48: Paragraphs (i)(2)(i) through (i)(2)(v), (i)(3), (i)(4)(i) through (i)(4)(v), (i)(5), (j), and (l)(1) of this AD.

Conclusion

We have carefully reviewed the available data, including the comments that have been received, and determined that air safety and the public interest require adopting the AD with the changes described previously. We have

determined that these changes will neither increase the economic burden on any operator nor increase the scope of the AD.

Explanation of Change to Costs of Compliance

Since issuance of the NPRM (75 FR 6154, February 8, 2010), we have increased the labor rate used in the Costs of Compliance from \$80 per work-hour to \$85 per work-hour. The Costs of Compliance information, below, reflects this increase in the specified hourly labor rate.

Costs of Compliance

There are about 975 airplanes of the affected design in the worldwide fleet. The following table provides the estimated costs for U.S. operators to comply with this AD.

ESTIMATED COSTS

Action	Work hours	Average labor rate per hour	Parts	Cost per airplane	Number of U.S.-registered airplanes	Fleet cost
Repetitive inspections of STA 1809.5 (required by AD 2006–24–04 (71 FR 68432, November 27, 2006)).	12	\$85	None	\$1,020 per inspection cycle.	354	\$361,080 per inspection cycle.
Inspection of inner chord (new action)	2	85	None	\$170 per inspection cycle.	354	\$60,180 per inspection cycle.

Authority for This Rulemaking

Title 49 of the United States Code specifies the FAA's authority to issue rules on aviation safety. Subtitle I, Section 106, describes the authority of the FAA Administrator. Subtitle VII, Aviation Programs, describes in more detail the scope of the Agency's authority.

We are issuing this rulemaking under the authority described in Subtitle VII, Part A, Subpart III, Section 44701, "General requirements." Under that section, Congress charges the FAA with promoting safe flight of civil aircraft in air commerce by prescribing regulations for practices, methods, and procedures the Administrator finds necessary for safety in air commerce. This regulation is within the scope of that authority because it addresses an unsafe condition that is likely to exist or develop on products identified in this rulemaking action.

Regulatory Findings

We have determined that this AD will not have federalism implications under Executive Order 13132. This AD will not have a substantial direct effect on the States, on the relationship between the national government and the States,

or on the distribution of power and responsibilities among the various levels of government.

For the reasons discussed above, I certify that this AD:

- (1) Is not a "significant regulatory action" under Executive Order 12866;
- (2) Is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and
- (3) Will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act.

We prepared a regulatory evaluation of the estimated costs to comply with this AD and placed it in the AD docket. See the **ADDRESSES** section for a location to examine the regulatory evaluation.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, under the authority delegated to me by the Administrator, the FAA amends 14 CFR part 39 as follows:

PART 39—AIRWORTHINESS DIRECTIVES

- 1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

- 2. The Federal Aviation Administration (FAA) amends § 39.13 by removing Amendment 39–14833 (71 FR 68432, November 27, 2006) and by adding the following new airworthiness directive (AD):

2011–14–02 The Boeing Company:
Amendment 39–16737. Docket No. FAA–2010–0033; Directorate Identifier 2009–NM–099–AD.

Effective Date

- (a) This AD becomes effective November 16, 2011.

Affected ADs

- (b) This AD supersedes AD 2006–24–04, Amendment 39–14833, (71 FR 68432, November 27, 2006).

Applicability

- (c) This AD applies to all The Boeing Company Model 767–200, –300, –300F, and –400ER series airplanes, certificated in any category.

Subject

(d) Air Transport Association (ATA) of America Code 53: Fuselage.

Unsafe Condition

(e) This AD results from reported fatigue cracking in the vertical inner chord and the forward outer chord while doing the detailed inspection of the horizontal inner chord at STA 1809.5. The Federal Aviation Administration is issuing this AD to detect and correct fatigue cracking in the bulkhead structure at STA 1809.5 and the vertical inner chord at STA 1809.5, which could result in failure of the bulkhead structure for carrying the flight loads of the horizontal stabilizer, and consequent loss of controllability of the airplane.

Compliance

(f) You are responsible for having the actions required by this AD performed within the compliance times specified, unless the actions have already been done.

Restatement of Requirements of AD 2006–24–04 (71 FR 68432, November 27, 2006), With Updated Service Information

Repetitive Inspections and Corrective Actions

(g) Before the accumulation of 15,000 total flight cycles, or within 3,000 flight cycles after January 2, 2007 (the effective date of AD 2006–24–04 (71 FR 68432, November 27, 2006)), whichever is later: Do the detailed and high frequency eddy current (HFEC) inspections for cracking as specified in Parts 1, 2, 3, and 4 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767–53A0131, dated March 30, 2006; or Revision 1, dated March 12, 2009; and do all corrective actions before further flight; by accomplishing all the actions specified in the Accomplishment Instructions of Boeing Alert Service Bulletin 767–53A0131, dated March 30, 2006; or Revision 1, dated March 12, 2009; except as provided by paragraph (h) of this AD. After the effective date of this AD, use only Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009. Repeat the inspections thereafter at intervals not to exceed 6,000 flight cycles. Accomplishing the corrective action for the inspections specified in Part 1, 2, 3, or 4, as applicable, of Boeing Alert Service Bulletin 767–53A0131, dated March 30, 2006; or Revision 1, dated March 12, 2009; as applicable; terminates the repetitive inspections for that area only.

Exceptions to Service Bulletin

(h) If any cracking is found in the skin or in any structure other than the forward outer chord, horizontal inner chord, or vertical inner chord during any inspection required by paragraph (g) or (k) of this AD, and Boeing Alert Service Bulletin 767–53A0131, dated March 30, 2006; or Revision 1, dated March 12, 2009; specifies to contact Boeing for appropriate action: Before further flight, repair the cracking using a method approved in accordance with the procedures specified in paragraph (n) of this AD; except that repairing or replacing stringers in accordance with Subject 53–00–03—Fuselage Stringers, of the applicable SRM identified in table 1 of this AD is an acceptable method of compliance for those specific actions required by this AD.

Optional Terminating Action for the Repetitive Inspections Required by Paragraph (g) of This AD

(i) For airplanes on which no cracking is found during the most recent detailed and HFEC inspections for a specified area as required by paragraph (g) of this AD: Paragraphs (i)(1) through (i)(5) of this AD provide optional terminating action for the repetitive inspections required by paragraph (g) of this AD for the specified area only.

(1) Modification of a specified area in accordance with a method approved by the Manager, Seattle Aircraft Certification Office (ACO), FAA, terminates the repetitive inspections required by paragraph (g) of this AD for that area only.

(2) Modification of a forward outer chord in accordance with the procedures specified in paragraphs (i)(2)(i) through (i)(2)(v) of this AD, as applicable, terminates the repetitive inspections required by paragraph (g) of this AD for that area only.

(i) For Model 767–200 series airplanes: Steps 4.A through 4.C and Steps 4.G through 4.P of Repair 9, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–200 Structural Repair Manual (SRM), Document D634T201.

(ii) For Model 767–300 series airplanes: Steps 4.A through 4.C and Steps 4.G through 4.P of Repair 9, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–300 SRM, Document D634T210.

(iii) For Model 767–300F series airplanes: Steps 4.A through 4.C and Steps 4.G through 4.P of Repair 9, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–300F SRM, Document D634T215.

(iv) For Model 767–300BCF series airplanes: Steps 4.A through 4.C and Steps 4.G through 4.P of Repair 9, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–300BCF SRM, Document D634T235.

(v) For Model 767–400 series airplanes: Steps 4.A through 4.C and Steps 4.G through 4.P of Repair 9, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–400 SRM, Document D634T225.

(3) Modification of a forward outer chord in accordance with Steps 4.A through 4.C and 4.G through 4.P of Repair 9 of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the applicable SRM identified in table 1 of this AD also terminates the repetitive inspections required by paragraph (g) of this AD for that area.

(4) Modification of a horizontal inner chord in accordance with the procedures specified in paragraphs (i)(4)(i) through (i)(4)(v) of this AD, as applicable, terminates the repetitive inspections required by paragraph (g) of this AD for that area.

(i) For Model 767–200 series airplanes: Steps 4.A, 4.B, and 4.F through 4.P of Repair 10, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–200 SRM, Document D634T201.

(ii) For Model 767–300 series airplanes: Steps 4.A, 4.B, and 4.F through 4.P of Repair 10, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–300 SRM, Document D634T210.

(iii) For Model 767–300F series airplanes: Steps 4.A, 4.B, and 4.F through 4.P of Repair 10, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–300F SRM, Document D634T215.

(iv) For Model 767–300BCF series airplanes: Steps 4.A, 4.B, and 4.F through 4.P of Repair 10, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–300BCF SRM, Document D634T235.

(v) For Model 767–400 series airplanes: Steps 4.A, 4.B, and 4.F through 4.P of Repair 10, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–400 SRM, Document D634T225.

(5) Modification of a horizontal inner chord in accordance with Steps 4.A, 4.B, and 4.F through 4.P of Repair 10 of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the applicable SRM identified in Table 1 of this AD also terminates the repetitive inspections required by paragraph (g) of this AD for that area.

TABLE 1—REVISED SRMS

SRM	Revision	Date
Boeing 767–200 SRM, Document D634T201	105	December 15, 2010.
Boeing 767–300 SRM, Document D634T210	85	December 15, 2010.
Boeing 767–300F SRM, Document D634T215	49	December 15, 2010.
Boeing 767–300BCF SRM, Document D634T235	9	December 15, 2010.
Boeing 767–400 SRM, Document D634T225	32	December 15, 2010.

Note 1: We have reformatted paragraph (i) of this AD to differentiate the methods of compliance specified in that paragraph.

Credit for Previously Accomplished Repairs

(j) Repair of a forward outer chord done before January 2, 2007, in accordance with Repair 9, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–200 SRM, Document D634T201; Boeing 767–300 SRM, Document D634T210; Boeing 767–300F SRM, Document D634T215; Boeing 767–300BCF SRM, Document D634T235; or Boeing 767–400 SRM, Document D634T225; as applicable; is acceptable for compliance with the requirements of paragraph (g) of this AD for that area only. Repair of a horizontal inner chord before January 2, 2007, in accordance with Repair 10, dated April 15, 2006, of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the Boeing 767–200 SRM, Document D634T201; Boeing 767–300 SRM, Document D634T210; Boeing 767–300F SRM, Document D634T215; Boeing 767–300BCF SRM, Document D634T235; or Boeing 767–400 SRM, Document D634T225; as applicable; is acceptable for compliance with the terminating requirements of paragraph (g) of this AD for that area only.

New Requirements of This AD

Inspections

(k) At the later of the times specified in paragraphs (k)(1) and (k)(2) of this AD: Do the detailed and HFEC inspections for cracking as specified in Parts 5 and 6 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009; and do all applicable corrective actions by accomplishing all the actions specified in the Accomplishment Instructions of Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009; except as provided by paragraph (h) of this AD. Do all applicable corrective actions before further flight. Repeat the inspections thereafter at intervals not to exceed 6,000 flight cycles. Accomplishing the corrective action for the inspections specified in Part 5 or 6 of the Accomplishment Instructions of Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009, as applicable, terminates the repetitive inspections for that area only.

(1) Before the accumulation of 15,000 total flight cycles or within 6,000 flight cycles after the inspection required by paragraph (g) of this AD, whichever occurs later.

(2) Within 30 days after the effective date of this AD.

Optional Terminating Action for the Repetitive Inspections Required by Paragraph (k) of This AD

(l) For airplanes on which no cracking is found during the most recent detailed and HFEC inspections for a specified area, as required by paragraph (k) of this AD: Paragraphs (l)(1) and (l)(2) of this AD provide optional terminating action for the repetitive inspections required by paragraph (k) of this AD for that area only. After the effective date of this AD, only the applicable SRM identified in table 1 of this AD or a method approved by the Manager, Seattle ACO, may be used.

(1) Modify the specified area in accordance with Steps 4.A through 4.C and 4.G through 4.Q of Repair 11 of Subject 53–80–08, Fuselage Bulkheads—Section 48, of the applicable SRM identified in table 1 or table 2 of this AD, except as provided by paragraph (m) of this AD.

(2) Modify the specified area in accordance with a method approved by the Manager, Seattle ACO.

TABLE 2—PREVIOUS SRMS

Steps—	Dated—	Of—
4.A through 4.C and 4.G through 4.Q of Repair 11 of Subject 53–80–08.	August 15, 2008	Boeing 767–200 SRM, Document D634T201.
4.A through 4.C and 4.G through 4.Q of Repair 11 of Subject 53–80–08.	August 15, 2008	Boeing 767–300 SRM, Document D634T210.
4.A through 4.C and 4.G through 4.Q of Repair 11 of Subject 53–80–08.	August 15, 2008	Boeing 767–300F SRM, Document D634T215.
4.A through 4.C and 4.G through 4.Q of Repair 11 of Subject 53–80–08.	August 15, 2008	Boeing 767–300BCF SRM, Document D634T235.
4.A through 4.C and 4.G through 4.Q of Repair 11 of Subject 53–80–08.	August 15, 2008	Boeing 767–400 SRM, Document D634T225.

Note 2: We have reformatted paragraph (l) of this AD to differentiate the methods of compliance specified in that paragraph.

Exception to SRM Modification Specified in Paragraph (l) of This AD

(m) If, during accomplishment of any modification in accordance with paragraph (l) of this AD, any cracking is found and the applicable SRM referenced in paragraph (l) of this AD specifies to contact Boeing for appropriate action: Before further flight, repair the cracking in accordance with a method approved by the Manager, Seattle ACO. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

Alternative Methods of Compliance (AMOCs)

(n)(1) The Manager, Seattle ACO, has the authority to approve AMOCs for this AD, if requested using the procedures found in 14 CFR 39.19. In accordance with 14 CFR 39.19, send your request to your principal inspector or local Flight Standards District Office, as appropriate. If sending information directly to the manager of the ACO, send it to the

attention of the person identified in the Related Information section of this AD. Or, e-mail information to 9-ANM-Seattle-ACO-AMOC-Requests@faa.gov.

(2) Before using any approved AMOC, notify your appropriate principal inspector, or lacking a principal inspector, the manager of the local flight standards district office/certificate holding district office.

(3) An AMOC that provides an acceptable level of safety may be used for any repair required by this AD, if it is approved by the Boeing Commercial Airplanes Organization Designation Authorization (ODA) that has been authorized by the Manager, Seattle ACO, to make those findings. For a repair method to be approved, the repair must meet the certification basis of the airplane, and the approval must specifically refer to this AD.

(4) AMOCs approved previously in accordance with AD 2006–24–04 (71 FR 68432, November 27, 2006) are approved as AMOCs for the corresponding provisions of this AD.

Material Incorporated by Reference

(o) You must use the service information specified in paragraph (o)(1) of this AD to do the actions required by this AD, unless the

AD specifies otherwise. If you accomplish the optional actions specified by this AD, you must use the service information specified in paragraph (o)(2) of this AD to perform those actions, unless the AD specifies otherwise. The Director of the Federal Register approved the incorporation by reference (IBR) under 5 U.S.C. 552(a) and 1 CFR part 51 of the following service information on November 16, 2011:

(1) Boeing Alert Service Bulletin 767–53A0131, Revision 1, dated March 12, 2009.

(2) Subject 53–00–03, Fuselage Stringers, of Chapter 53, Fuselage; or Subject 53–80–08, Fuselage Bulkheads—Section 48, Repair 9—Station 1809.5 Bulkhead—Forward Outer Chord Repair Between S–4 to S–8, Repair 10—Station 1809.5 Bulkhead—Horizontal Inner Chord Repair at Approximately WL 257 and BL 28, or Repair 11—Station 1809.5 Bulkhead—Vertical Inner Chord Repair at Approximately WL 256 and BL 30, as applicable; of Chapter 53, Fuselage; as applicable; of the applicable Structural Repair Manual (SRM) specified in paragraphs (o)(2)(i) through (o)(2)(v) of this AD.

(i) Boeing 767–200 SRM, Document D634T201, Revision 105, dated December 15, 2010. Only the transmittal letter, dated

December 15, 2010, of this document contains the revision level of the document.

(ii) Boeing 767–300 SRM, Document D634T210, Revision 85, dated December 15, 2010. Only page 1 of the transmittal letter, dated December 15, 2010, of this document contains the revision level of the document.

(iii) Boeing 767–300F SRM, Document D634T215, Revision 49, dated December 15, 2010. Only page 1 of the transmittal letter, dated December 15, 2010, of this document contains the revision level of the document.

(iv) Boeing 767–300BCF SRM, Document D634T235, Revision 9, dated December 15, 2010. Only page 1 of the transmittal letter, dated December 15, 2010, of this document contains the revision level of the document.

(v) Boeing 767–400 SRM, Document D634T225, Revision 32, dated December 15, 2010. Only page 1 of the transmittal letter, dated December 15, 2010, of this document contains the revision level of the document.

(3) For service information identified in this AD, contact Boeing Commercial Airplanes, Attention: Data & Services Management, P.O. Box 3707, MC 2H–65, Seattle, Washington 98124–2207; telephone 206–544–5000, extension 1; fax 206–766–5680; e-mail me.boecom@boeing.com; Internet <https://www.myboeingfleet.com>.

(4) You may review copies of the service information at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington. For information on the availability of this material at the FAA, call 425–227–1221.

(5) You may also review copies of the service information that is incorporated by reference at the National Archives and Records Administration (NARA). For information on the availability of this material at NARA, call 202–741–6030, or go to: http://www.archives.gov/federal_register/code_of_federal_regulations/ibr_locations.html.

Issued in Renton, Washington, on September 23, 2011.

Ali Bahrami,

Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 2011–25618 Filed 10–11–11; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. FAA–2011–0389; Directorate Identifier 2007–NM–189–AD; Amendment 39–16769; AD 2011–17–05]

RIN 2120–AA64

Airworthiness Directives; Airbus Model A300 B2–1C, A300 B2–203, A300 B2K–3C, A300–B4–103, A300 B4–203, and A300 B4–2C Airplanes

AGENCY: Federal Aviation Administration (FAA), Department of Transportation (DOT).

ACTION: Final rule.

SUMMARY: We are superseding an existing airworthiness directive (AD) that applies to the products listed above. This AD results from mandatory continuing airworthiness information (MCAI) originated by an aviation authority of another country to identify and correct an unsafe condition on an aviation product. The MCAI describes the unsafe condition as:

* * * [C]racks * * * in sections 13 to 18 of the fuselage between rivets of longitudinal lap joints between frames 18 and 80 which could affect the structural integrity of the fuselage if not corrected.

* * * * *

We are issuing this AD to require actions to correct the unsafe condition on these products.

DATES: This AD becomes effective November 16, 2011.

The Director of the Federal Register approved the incorporation by reference of certain publications listed in this AD as of November 16, 2011.

ADDRESSES: You may examine the AD docket on the Internet at <http://www.regulations.gov> or in person at the U.S. Department of Transportation, Docket Operations, M–30, West Building Ground Floor, Room W12–140, 1200 New Jersey Avenue, SE., Washington, DC.

FOR FURTHER INFORMATION CONTACT: Dan Rodina, Aerospace Engineer, International Branch, ANM–116, Transport Airplane Directorate, FAA, 1601 Lind Avenue, SW., Renton, Washington 98057–3356; telephone (425) 227–2125; fax (425) 227–1149.

SUPPLEMENTARY INFORMATION:

Discussion

We issued a notice of proposed rulemaking (NPRM) to amend 14 CFR part 39 to include an AD that would apply to the specified products. That NPRM was published in the **Federal Register** on May 10, 2011 (76 FR 26962), and proposed to supersede AD 90–01–10, Amendment 39–6448 (55 FR 261, January 4, 1990). That NPRM proposed to correct an unsafe condition for the specified products. The MCAI states:

This Airworthiness Directive (AD) is issued in order to prevent cracks development in sections 13 to 18 of the fuselage between rivets of longitudinal lap joints between frames 18 and 80 which could affect the structural integrity of the fuselage if not corrected.

This new AD:

- Retains the requirements of DGAC AD 1989–061–092(B)R4 [which corresponds to FAA AD 90–01–10 (55 FR 261, January 4, 1990)], which is cancelled;
- Takes into account a new inspection program as detailed in AIRBUS Service

Bulletins (SB) A300–53–0211 Revision 7, which will allow A300 aircraft to reach the Limit of Validity (LOV).

This AD has been republished to correctly refer to SB A300–53–0211 in Note 2 of the Compliance section.

The inspection program consists of repetitive detailed inspections for disbonding and cracking of the fuselage inner doubler; eddy current and ultrasonic inspections of the fuselage longitudinal lap joints for cracking; and repair if necessary (*i.e.*, repairing any cracking or disbonding, or contacting Airbus for repair instructions and doing the repair). You may obtain further information by examining the MCAI in the AD docket.

Comments

We gave the public the opportunity to participate in developing this AD. We received no comments on the NPRM (76 FR 26962, May 10, 2011) or on the determination of the cost to the public.

Clarification of Service Bulletin References

Paragraphs (h)(1) and (j)(1) of the NPRM (76 FR 26962, May 10, 2011) refer to Airbus Mandatory Service Bulletin A300–53–0211 as the service information for airplanes on which an inspection of the longitudinal lap joints has been done. For clarity, we have revised the paragraphs to refer to the latest service bulletin revision and, therefore, we have revised paragraphs (h)(1) and (j)(1) of this AD to refer to Airbus Mandatory Service Bulletin A300–53–0211, Revision 07, dated December 1, 2006.

Paragraphs (l)(1), (1)(2), (m)(1), (m)(2), (n)(1), (n)(2), (o)(1), and (o)(2) of the NPRM (76 FR 26962, May 10, 2011) refer to Airbus Service Bulletin A300–53–229 for the definition of “major” and “minor” disbonding. For clarity, we have revised the paragraphs to refer to the latest service bulletin revision and, therefore, we have revised paragraphs (l)(1), (1)(2), (m)(1), (m)(2), (n)(1), (n)(2), (o)(1), and (o)(2) of this AD to refer to Airbus Service Bulletin A300–53–229, Revision 5, dated April 8, 1997.

Conclusion

We reviewed the available data and determined that air safety and the public interest require adopting the AD with the changes described previously. We determined that these changes will not increase the economic burden on any operator or increase the scope of the AD.