

FAA's Conclusions

These airplane models are manufactured in France and are type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the DGAC has kept us informed of the situation described above. We have examined the findings of the DGAC, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the applicable service bulletin described previously.

Cost Impact

The FAA estimates that 468 airplanes of U.S. registry would be affected by this proposed AD.

It would take between 10 and 22 work hours per airplane to accomplish the proposed modification, at an average labor rate of \$65 per work hour. Required parts would cost between \$670 and \$5750 per airplane. Based on these figures, the cost impact of the modification proposed by this AD is estimated to be between \$617,760 and \$3,360,240, or between \$1,320 and \$7,180 per airplane.

Should an operator be required to do the replacement of the high-level sensors, it would take about 80 work hours, at an average labor rate of \$65 per work hour. Required parts would be free of charge. Based on these figures, the cost impact of the replacement proposed by this AD is estimated to be \$5,200 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Airbus: Docket 2002–NM–153–AD.

Applicability: Model A319 and A320 series airplanes; certificated in any category; as listed in Airbus Service Bulletin A320–28–1087, Revision 02, dated June 10, 2003; and Airbus Service Bulletin A320–28–1086, Revision 01, dated October 23, 2002.

Compliance: Required as indicated, unless accomplished previously.

To prevent overheating of the fuel probes due to a short circuit, and fuel leakage due to inadequate expansion of the area within the additional center fuel tank, which could result in fuel vapors or fuel contacting an ignition source and/or consequent fire/explosion in the center fuel tank, accomplish the following:

Modification/Replacement

(a) Within 4,000 flight hours or 30 months after the effective date of this AD, whichever is first: Do the applicable actions specified in paragraphs (a)(1) and (a)(2) of this AD. Accomplishment of the modification before the effective date of this AD per Airbus Service Bulletin A320–28–1087, dated July 17, 2001; or Revision 01, dated March 3, 2003; or accomplishment of the replacement before the effective date of this AD per Airbus Service Bulletin A320–28–1086, dated November 30, 1999; as applicable; is considered acceptable for compliance with the corresponding action specified in paragraph (a)(1) or (a)(2) of this AD.

(1) For airplanes defined in Airbus Service Bulletin A320–28–1087, Revision 02, dated June 10, 2003: Modify the wiring of the fuel quantity indicating probes of the center and wing fuel tanks by doing all the actions specified in paragraphs 3.A. through 3.D. (including operational testing and any applicable repair) of the Accomplishment Instructions of the service bulletin. Do the actions per the service bulletin. Any applicable repair must be done before further flight.

(2) For airplanes defined in Airbus Service Bulletin A320–28–1086, Revision 01, dated October 23, 2002: Prior to or concurrent with accomplishment of paragraph (a)(1) of this AD, replace the high-level sensors of the additional center fuel tanks by doing all the actions specified in paragraphs 3.A through 3.D. (including operational testing and any applicable repair) of the Accomplishment Instructions of the service bulletin. Do the actions per the service bulletin. Any applicable repair must be done before further flight.

Alternative Methods of Compliance

(b) In accordance with 14 CFR 39.19, the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate, is authorized to approve alternative methods of compliance for this AD.

Note 1: The subject of this AD is addressed in French airworthiness directive 2002–220(B) R1, dated October 15, 2003.

Issued in Renton, Washington, on November 10, 2003.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03–28609 Filed 11–14–03; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2002–NM–120–AD]

RIN 2120–AA64

Airworthiness Directives; Bombardier Model DHC–8–401 and –402 Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Bombardier Model DHC-8-401 and -402 airplanes. This proposal would require modifying the wiring of the rudder trim switch, inspecting all wiring on the back of the aileron/rudder trim control panel for chafing, and replacing any chafed wiring with new wiring. This action is necessary to prevent a short circuit on the aileron/rudder trim control panel that could cause a runaway condition of the rudder trim actuator, which could result in reduced controllability of the airplane. This action is intended to address the identified unsafe condition.

DATES: Comments must be received by December 17, 2003.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2002-NM-120-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227-1232. Comments may also be sent via the Internet using the following address: 9-anm-nprmcomment@faa.gov. Comments sent via fax or the Internet must contain "Docket No. 2002-NM-120-AD" in the subject line and need not be submitted in triplicate. Comments sent via the Internet as attached electronic files must be formatted in Microsoft Word 97 or 2000 or ASCII text.

The service information referenced in the proposed rule may be obtained from Bombardier, Inc., Bombardier Regional Aircraft Division, 123 Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York.

FOR FURTHER INFORMATION CONTACT: Douglas Wagner, Aerospace Engineer, Systems and Flight Test Branch, ANE-172, FAA, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York 11581; telephone (516) 256-7506; fax (516) 568-2716.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this action may be changed in light of the comments received.

Submit comments using the following format:

- Organize comments issue-by-issue. For example, discuss a request to change the compliance time and a request to change the service bulletin reference as two separate issues.
- For each issue, state what specific change to the proposed AD is being requested.
- Include justification (*e.g.*, reasons or data) for each request.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this action must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 2002-NM-120-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 2002-NM-120-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

Transport Canada Civil Aviation (TCCA), which is the airworthiness authority for Canada, notified the FAA that an unsafe condition may exist on certain Bombardier Model DHC-8-401 and -402 airplanes. TCCA advises that an incident of runaway of the rudder trim actuator occurred immediately following take-off. Investigation revealed a discrepancy in the wiring of

the rudder trim control switch on the aileron/rudder trim control panel. This resulted in the rudder trim control switch being constantly enabled. In the event of a short-circuit of a wire, this condition, if not corrected, could result in a runaway condition of the rudder trim actuator and consequent reduced controllability of the airplane.

Explanation of Relevant Service Information

Bombardier has issued Alert Service Bulletin A84-27-13, Revision "B," dated January 12, 2002. That service bulletin describes procedures for modifying the wiring of the rudder trim switch, and performing a one-time general visual inspection of all wiring on the back of the aileron/rudder trim control panel for chafing, and replacement of any chafed wiring with new wiring. Accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition. TCCA classified this service bulletin as mandatory and issued Canadian airworthiness directive CF-2002-15, dated February 20, 2002, to ensure the continued airworthiness of these airplanes in Canada.

FAA's Conclusions

This airplane model is manufactured in Canada and is type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, TCCA has kept the FAA informed of the situation described above. The FAA has examined the findings of TCCA, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed rule would require accomplishment of the actions specified in the service bulletin described previously.

Cost Impact

The FAA estimates that 12 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 1 work hour per airplane to accomplish the proposed actions, and that the average labor rate is \$65 per

work hour. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be \$780, or \$65 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted. The cost impact figures discussed in AD rulemaking actions represent only the time necessary to perform the specific actions actually required by the AD. These figures typically do not include incidental costs, such as the time required to gain access and close up, planning time, or time necessitated by other administrative actions.

Regulatory Impact

The regulations proposed herein would not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this proposal would not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this proposed regulation (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket. A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

Bombardier, Inc. (Formerly de Havilland, Inc.): Docket 2002–NM–120–AD.

Applicability: Model DHC–8–401 and –402 airplanes; certificated in any category; serial numbers 4005, 4006, 4008 through 4016 inclusive, and 4018 through 4058 inclusive.

Compliance: Required as indicated, unless accomplished previously.

To prevent a short circuit on the aileron/rudder trim control panel that could cause a runaway condition of the rudder trim actuator, which could result in reduced controllability of the airplane, accomplish the following:

Modification, Inspection, and Corrective Action

(a) Within 90 days after the effective date of this AD, do the actions in paragraphs (a)(1) and (a)(2) of this AD, per the Accomplishment Instructions of Bombardier Alert Service Bulletin A84–27–13, Revision "B," dated January 12, 2002.

(1) Modify the wiring of the rudder trim switch.

(2) Before further flight after accomplishing the modification required by paragraph (a)(1) of this AD: Perform a one-time general visual inspection of all wiring on the back of the aileron/rudder trim control panel for chafing. Before further flight, replace any chafed wiring with new wiring.

Note 1: For the purposes of this AD, a general visual inspection is defined as: "A visual examination of an interior or exterior area, installation, or assembly to detect obvious damage, failure, or irregularity. This level of inspection is made from within touching distance unless otherwise specified. A mirror may be necessary to enhance visual access to all exposed surfaces in the inspection area. This level of inspection is made under normally available lighting conditions such as daylight, hangar lighting, flashlight, or droplight and may require removal or opening of access panels or doors. Stands, ladders, or platforms may be required to gain proximity to the area being checked."

Previously Accomplished Actions

(b) Modifications and inspections accomplished before the effective date of this AD per Bombardier Alert Service Bulletin A84–27–13, Revision "A," dated January 9, 2002, are acceptable for compliance with the corresponding actions required by paragraph (a) of this AD.

Parts Installation

(c) As of the effective date of this AD, no person may install aileron/rudder trim control panel having part number 82410608–005 on any airplane, unless the control panel has been modified and inspected per the requirements of this AD.

Alternative Methods of Compliance

(d) In accordance with 14 CFR 39.19, the Manager, New York Aircraft Certification Office (ACO), FAA, is authorized to approve alternative methods of compliance for this AD.

Note 2: The subject of this AD is addressed in Canadian airworthiness directive CF–2002–15, dated February 20, 2002.

Issued in Renton, Washington, on November 10, 2003.

Kalene C. Yanamura,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 03–28608 Filed 11–14–03; 8:45 am]

BILLING CODE 4910–13–P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 2001–NM–133–AD]

RIN 2120–AA64

Airworthiness Directives; McDonnell Douglas Model DC–8–70 and –70F Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain McDonnell Douglas Model DC–8–70 and –70F series airplanes. This proposal would require repetitive inspections for cracking of the lower cargo doorjamb corners, and corrective action if necessary. For certain airplanes, this proposal would provide for optional terminating action for certain repetitive inspections. For certain other airplanes, this proposal would require modification of the lower cargo doorjamb corners. This action is necessary to detect and correct cracking in the lower cargo doorjamb corners, which could result in rapid decompression of the fuselage and consequent reduced structural integrity of the airplane. This action is intended to address the identified unsafe condition.

DATES: Comments must be received by January 2, 2004.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM–114, Attention: Rules Docket No. 2001–NM–133–AD, 1601 Lind Avenue, SW., Renton, Washington 98055–4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays. Comments may be submitted via fax to (425) 227–1232. Comments may also be sent via the Internet using the following address: *9-anm-*