

**DEPARTMENT OF TRANSPORTATION****Federal Aviation Administration****14 CFR Part 39**

[Docket No. 99-NM-344-AD; Amendment 39-11589; AD 2000-04-07]

RIN 2120-AA64

**Airworthiness Directives; British Aerospace BAe Model ATP Airplanes**

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD), applicable to all British Aerospace BAe Model ATP airplanes, that requires a one-time detailed visual inspection to detect incorrect installation or discrepancies (damage, bending, overheating, discoloration) of the circuit breaker and the cable terminations of the circuit breaker of the engine de-ice panel. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to prevent failure of the engine intake de-icing system, which could result in loss of engine intake de-icing capability, accretion of ice in the intake duct, ice ingestion, and consequent engine flameout.

**DATES:** Effective March 30, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of March 30, 2000.

**ADDRESSES:** The service information referenced in this AD may be obtained from British Aerospace Regional Aircraft American Support, 13850 Mclearen Road, Herndon, Virginia 20171. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**FOR FURTHER INFORMATION CONTACT:** Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

**SUPPLEMENTARY INFORMATION:** A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to all British

Aerospace BAe Model ATP airplanes was published in the **Federal Register** on December 7, 1999 (64 FR 68296). That action proposed to require a one-time detailed visual inspection to detect incorrect installation or discrepancies (damage, bending, overheating, discoloration) of the circuit breaker and the cable terminations of the circuit breaker of the engine de-ice panel.

**Comments**

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

**Conclusion**

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

**Cost Impact**

The FAA estimates that 10 airplanes of U.S. registry will be affected by this AD, that it will take approximately 2 work hours per airplane to accomplish the required inspection, and that the average labor rate is \$60 per work hour. Based on these figures, the cost impact of the AD on U.S. operators is estimated to be \$1,200, or \$120 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

**Regulatory Impact**

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules

Docket at the location provided under the caption **ADDRESSES**.

**List of Subjects in 14 CFR Part 39**

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

**Adoption of the Amendment**

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

**PART 39—AIRWORTHINESS DIRECTIVES**

1. The authority citation for part 39 continues to read as follows:

**Authority:** 49 U.S.C. 106(g), 40113, 44701.

**§ 39.13 [Amended]**

2. Section 39.13 is amended by adding the following new airworthiness directive:

**2000-04-07 British Aerospace Regional Aircraft** [Formerly Jetstream Aircraft Limited; British Aerospace (Commercial Aircraft) Limited]: Amendment 39-11589. Docket 99-NM-344-AD.

**Applicability:** All Model ATP airplanes, certificated in any category.

**Note 1:** This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

**Compliance:** Required as indicated, unless accomplished previously.

To prevent failure of the engine intake de-icing system, which could result in loss of engine intake de-icing capability, accretion of ice in the intake duct, ice ingestion, and consequent engine flameout, accomplish the following:

**One-Time Inspection**

(a) Within 3 months after the effective date of this AD: Perform a one-time detailed visual inspection to detect incorrect installation or discrepancies (damage, bending, overheating, discoloration) of the circuit breaker and the cable terminations of the circuit breaker of the engine de-ice panel, in accordance with Part 5 of the Accomplishment Instructions of British Aerospace Service Bulletin ATP-30-52, Revision 1, dated June 12, 1998. If any incorrect installation or discrepancy is

detected, prior to further flight, repair it in accordance with the service bulletin.

**Note 2:** For the purposes of this AD, a detailed visual inspection is defined as: "An intensive visual examination of a specific structural area, system, installation, or assembly to detect damage, failure, or irregularity. Available lighting is normally supplemented with a direct source of good lighting at intensity deemed appropriate by the inspector. Inspection aids such as mirror, magnifying lenses, etc., may be used. Surface cleaning and elaborate access procedures may be required."

#### Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

**Note 3:** Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the Manager, International Branch, ANM-116.

#### Special Flight Permits

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

#### Incorporation by Reference

(d) The inspection and repair shall be done in accordance with British Aerospace Service Bulletin ATP-30-52, Revision 1, dated June 12, 1998. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from British Aerospace Regional Aircraft American Support, 13850 Mclearen Road, Herndon, Virginia 20171. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

**Note 4:** The subject of this AD is addressed in British airworthiness directive 007-01-98.

(e) This amendment becomes effective on March 30, 2000.

Issued in Renton, Washington, on February 15, 2000.

**Donald L. Riggins,**

*Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.*  
[FR Doc. 00-4117 Filed 2-23-00; 8:45 am]

**BILLING CODE 4910-13-U**

## DEPARTMENT OF TRANSPORTATION

### Federal Aviation Administration

#### 14 CFR Part 39

[Docket No. 99-NM-352-AD; Amendment 39-11590; AD 2000-04-08]

RIN 2120-AA64

#### Airworthiness Directives; Boeing Model 737-200C Series Airplanes

**AGENCY:** Federal Aviation Administration, DOT.

**ACTION:** Final rule; request for comments.

**SUMMARY:** This amendment adopts a new airworthiness directive (AD) that is applicable to certain Boeing Model 737-200C series airplanes. This action requires repetitive inspections to detect cracking in the lower skin at the stringer 4R lap joint, and certain fuselage frames; and corrective actions, if necessary. This amendment also provides for optional terminating action for the repetitive inspections. This amendment is prompted by a report of a fractured frame located at body station (BS) 480. The actions specified in this AD are intended to detect and correct cracking in certain frames, which, in conjunction with multiple site cracking in the lower skin of the lap joint, could result in failure of certain lap joints, and consequent rapid decompression of the airplane fuselage.

**DATES:** Effective March 10, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of March 10, 2000.

Comments for inclusion in the Rules Docket must be received on or before April 24, 2000.

**ADDRESSES:** Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 99-NM-352-AD, 1601 Lind Avenue, SW., Renton, WA 98055-4056.

The service information referenced in this AD may be obtained from Boeing Commercial Airplane Group, PO Box 3707, Seattle, WA 98124-2207. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, WA; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

#### FOR FURTHER INFORMATION CONTACT:

James G. Rehrl, Aerospace Engineer, Airframe Branch, ANM-120S, FAA, Transport Airplane Directorate, Seattle

Aircraft Certification Office, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2783; fax (425) 227-1181.

**SUPPLEMENTARY INFORMATION:** The FAA has received a report indicating that a cracked fuselage frame was detected at body station (BS) 480, common to the stringer 5R integral shear tie stringer cutout on a Model 737-200C series airplane. Subsequent investigation revealed that the z-frame and integral-shear-tie at the stringer cutout were fractured. A crack also was detected in the S-5R and S-2R integral shear ties at the stringer cutout. A fractured frame at stringer 5R is a concern because it is adjacent to the stringer 4R lap joint, which is susceptible to multiple site cracking. Although the cracking was detected at BS 480 only, the frames at BS 500, 500A, 500B, and 520 have a similar design. Such cracking in these frames, in conjunction with multiple site cracking in the lower skin of the stringer 4R lap joint, could result in failure of the lap joint, and consequent rapid decompression of the airplane fuselage.

#### Explanation of Relevant Service Information

The FAA has reviewed and approved Boeing Alert Service Bulletin 737-53A1220, dated October 4, 1999, which describes procedures for a low frequency eddy current (LFEC) inspection to detect cracking in the lower skin at the stringer 4R lap joint between BS 460 and BS 540, and a detailed internal visual inspection to detect cracking in the frames at BS 480 through BS 520. The alert service bulletin also describes procedures for a preventative modification of the BS 480 frame.

#### Explanation of the Requirements of the Rule

Since an unsafe condition has been identified that is likely to exist or develop on other Model 737-200C series airplanes of the same type design, this AD is being issued to require repetitive inspections to detect cracking in the lower skin at the stringer 4R lap joint, and certain fuselage frames; and corrective actions, if necessary. This amendment also provides for optional terminating action for the repetitive inspections. The actions are required to be accomplished in accordance with the alert service bulletin described previously, except as discussed below.

#### Differences Between Service Bulletin and This AD

Operators should note that, although the alert service bulletin does not