

1. *Seats With Inflatable Lapbelts.* It must be shown that the inflatable lapbelt will deploy and provide protection under crash conditions where it is necessary to prevent serious head injury. The means of protection must take into consideration a range of stature from a two-year-old child to a ninety-fifth percentile male. The inflatable lapbelt must provide a consistent approach to energy absorption throughout that range. In addition, the following situations must be considered:

- a. The seat occupant is holding an infant.
- b. The seat occupant is a child in a child restraint device.
- c. The seat occupant is a child not using a child restraint device.
- d. The seat occupant is a pregnant woman.

2. The inflatable lapbelt must provide adequate protection for each occupant regardless of the number of occupants of the seat assembly, considering that unoccupied seats may have active seatbelts.

3. The design must prevent the inflatable lapbelt from being either incorrectly buckled or incorrectly installed such that the airbag would not properly deploy. Alternatively, it must be shown that such deployment is not hazardous to the occupant, and will provide the required head injury protection.

4. It must be shown that the inflatable lapbelt system is not susceptible to inadvertent deployment as a result of wear and tear, or inertial loads resulting from in-flight or ground maneuvers (including gusts and hard landings), likely to be experienced in service.

5. Deployment of the inflatable lapbelt must not introduce injury mechanisms to the seated occupant, or result in injuries that could impede rapid egress. This assessment should include an occupant who is in the brace position when it deploys and an occupant whose belt is loosely fastened.

6. It must be shown that an inadvertent deployment, that could cause injury to a standing or sitting person, is improbable.

7. It must be shown that inadvertent deployment of the inflatable lapbelt, during the most critical part of the flight, will either not cause a hazard to the airplane or is extremely improbable.

8. It must be shown that the inflatable lapbelt will not impede rapid egress of occupants 10 seconds after its deployment.

9. The system must be protected from lightning and HIRF. The threats specified in Special Condition No. 25-ANM-78 are incorporated by reference

for the purpose of measuring lightning and HIRF protection. For the purposes of complying with HIRF requirements, the inflatable lapbelt system is considered a "critical system" if its deployment could have a hazardous effect on the airplane; otherwise it is considered an "essential" system.

10. The inflatable lapbelt must function properly after loss of normal aircraft electrical power, and after a transverse separation of the fuselage at the most critical location. A separation at the location of the lapbelt does not have to be considered.

11. It must be shown that the inflatable lapbelt will not release hazardous quantities of gas or particulate matter into the cabin.

12. The inflatable lapbelt installation must be protected from the effects of fire such that no hazard to occupants will result.

13. There must be a means for a crewmember to verify the integrity of the inflatable lapbelt activation system prior to each flight or it must be demonstrated to reliably operate between inspection intervals.

Issued in Renton, Washington, on March 20, 2000.

Donald L. Riggan,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service, ANM-100.

[FR Doc. 00-7633 Filed 3-27-00; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NE-57-AD; Amendment 39-11632; AD 2000-05-22]

RIN 2120-AA64

Airworthiness Directives; CFM International CFM56-2, -2A, -2B, -3, -3B, and -3C Series Turbofan Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD) that is applicable to CFM International CFM56-2, -2A, -2B, -3, -3B, and -3C series turbofan engines. This amendment requires a one-time eddy current inspection (ECI) for cracks in the bolt holes of high pressure turbine (HPT) front rotating air seals. This amendment is prompted by reports of machining anomalies in a bolt hole that led to an HPT front rotating air seal

failure. The actions specified by this AD are intended to detect cracks in the bolt holes of HPT front rotating air seals, which can lead to an uncontained engine failure and damage to the aircraft.

DATES: Effective May 2, 2000.

The incorporation by reference of certain publications in this rule is approved by the Director of the Federal Register as of May 2, 2000.

ADDRESSES: The service information referenced in this AD may be obtained from CFM International, Technical Publications Department, 1 Neumann Way, Cincinnati, OH 45215; telephone (513) 552-2800, fax (513) 552-2816. This information may be examined at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA.

FOR FURTHER INFORMATION CONTACT: James Rosa, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive Park, Burlington, MA 01803-5299; telephone (781) 238-7152, fax (781) 238-7199.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to CFM International CFM56-2, -2A, -2B, -3, -3B, and -3C Series Turbofan Engines was published in the **Federal Register** on December 13, 1999 (64 FR 69248). That action proposed to require a one-time eddy current inspection (ECI) for cracks in the bolt holes of high pressure turbine (HPT) front rotating air seals. That action was prompted by reports of machining anomalies in a bolt hole that led to an HPT front rotating air seal failure. That condition, if not corrected could result in cracks in the bolt holes of HPT front rotating air seals, which can lead to an uncontained engine failure and damage to the aircraft.

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were received.

After careful review of the available data, the FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Economic Analysis

There are approximately 121 engines of the affected design in the worldwide fleet. The FAA estimates that 13 engines installed on aircraft of US registry will be affected by this AD, that it would take approximately 300 work hours per engine to accomplish the actions, and that the average labor rate is \$60 per work hour. Based on these figures, the

total cost impact of the AD on US operators is estimated to be \$234,000.

Regulatory Impact

The regulations adopted herein will not have a substantial direct effect on the States, on the relationship between the national Government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, it is determined that this final rule does not have federalism implications under Executive Order 13132.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

2000-05-22 CFM International:

Amendment 39-11632. Docket 99-NE-57-AD.

Applicability: CFM International (CFMI) CFM56-2, -2A, -2B, -3, -3B, and -3C series turbofan engines, installed on but not limited to McDonnell Douglas DC-8 series, Boeing 737 series, as well as Boeing E-3, E-6, and KC-135 (Military) series airplanes.

Note 1: This airworthiness directive (AD) applies to each engine identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the

requirements of this AD. For engines that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (c) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect cracks in the bolt holes of high pressure turbine (HPT) front rotating air seals, which can lead to an uncontained engine failure and damage to the aircraft, accomplish the following:

One-Time Eddy Current Inspections (ECI) Based Upon Engine Model and Thrust Ratings

(a) Perform a one-time ECI for cracks in the bolt holes of HPT front rotating air seals, part number 1282M72P03, and, if necessary, replace with serviceable parts, as follows:

CFM56-3 Series

(1) For CFM56-3-B1 engine nameplate models with HPT front rotating air seals listed by serial number (S/N) in paragraph 1.A(1), Effectivity, of CFMI CFM56-3/3B/3C Service Bulletin (SB) 72-922, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, and in accordance with the intervals listed in paragraph (a)(4)(i) or (a)(4)(ii) of this AD, as applicable.

(2) For CFM56-3B-2 models with maximum thrust limited to 20,100 or 18,500 pounds by the flight management computer (FMC) and aircraft flight manual (AFM), with HPT front rotating air seals listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-3/3B/3C SB 72-922, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, and in accordance with the intervals listed in paragraph (a)(4)(i) or (a)(4)(ii) of this AD, as applicable.

(3) For CFM56-3C-1 models with maximum thrust limited to 20,100 or 18,500 pounds by the FMC and AFM, with HPT front rotating air seals listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-3/3B/3C SB 72-922, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, and in accordance with the intervals listed in paragraph (a)(4)(i) or (a)(4)(ii), as applicable.

Compliance Times for (a)(1), (a)(2), and (a)(3)

(4) Use the following compliance times for the engine models listed in paragraphs (a)(1), (a)(2), and (a)(3) of this AD:

(i) For HPT front rotating air seals with less than 10,000 cycles since new (CSN) on the effective date of this AD, inspect at the next engine shop visit after accumulating 4,000 CSN, not to exceed 13,000 CSN.

(ii) For HPT front rotating air seals with 10,000 CSN or more on the effective date of

this AD, inspect at the next engine shop visit prior to accumulating 3,000 cycles-in-service (CIS) after the effective date of this AD, or prior to accumulating 20,000 CSN, whichever occurs first.

(5) For CFM56-3B-2 engine nameplate models, with HPT front rotating air seals listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-3/3B/3C SB 72-922, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, and in accordance with the intervals listed in paragraphs (a)(7)(i), or (a)(7)(ii) of this AD, as applicable.

(6) For CFM56-3C-1 models with maximum thrust limited to 22,100 pounds by the FMC and AFM, with HPT front rotating air seals listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-3/3B/3C SB 72-922, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, and in accordance with the intervals listed in paragraphs (a)(7)(i), or (a)(7)(ii) of this AD, as applicable.

Compliance Times for (a)(5) and (a)(6)

(7) Use the following compliance times for the engine models listed in paragraphs (a)(5) and (a)(6) of this AD:

(i) For HPT front rotating air seals with less than 9,800 CSN on the effective date of this AD, inspect at the next engine shop visit after accumulating 4,000 CSN, not to exceed 12,800 CSN.

(ii) For HPT front rotating air seals with 9,800 CSN or more on the effective date of this AD, inspect at the next engine shop visit prior to accumulating 3,000 CIS after the effective date of this AD, or prior to accumulating 15,800 CSN, whichever occurs first.

(8) For CFM56-3C-1 engine nameplate models, with HPT front rotating air seals listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-3/3B/3C SB 72-922, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, as follows:

(i) For HPT front rotating air seals with less than 9,100 CSN on the effective date of this AD, inspect at the next engine shop visit after accumulating 4,000 CSN, not to exceed 12,100 CSN.

(ii) For HPT front rotating air seals with 9,100 CSN or more on the effective date of this AD, inspect at the next engine shop visit prior to accumulating 3,000 CIS after the effective date of this AD, or prior to accumulating 15,100 CSN, whichever occurs first.

Uninstalled Parts

(9) Prior to installation in CFM56-3/3B/3C series engines, inspect uninstalled parts listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-3/3B/3C SB 72-922, dated November 12, 1999, in accordance with Paragraph 2, Accomplishment Instructions, of that SB.

CFM56-2 Series

(10) For CFM56-2 engine nameplate models, with HPT front rotating air seals listed by S/N in paragraph 1.A(1), Effectivity,

of CFMI CFM56-2 SB 72-869, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, as follows:

(i) For HPT front rotating air seals with less than 9,100 CSN on the effective date of this AD, inspect at the next engine shop visit after accumulating 4,000 CSN, not to exceed 10,100 CSN.

(ii) For HPT front rotating air seals with 9,100 CSN or more on the effective date of this AD, inspect at the next engine shop visit prior to accumulating 1,000 CIS after the effective date of this AD, or prior to accumulating 13,100 CSN, whichever occurs first.

Uninstalled Parts

(11) Prior to installation in CFM56-2 series engines, inspect uninstalled parts listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-2 SB 72-869, dated November 12, 1999, in accordance with Paragraph 2, Accomplishment Instructions, of that SB.

CFM56-2A Series

(12) For CFM56-2A engine nameplate models, with HPT front rotating air seals listed by S/N in paragraph 1.A(1), Effectivity, of CFM56-2A SB 72-470, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, after accumulating 3,000 CSN but before accumulating 6,000 CSN.

Uninstalled Parts

(13) Prior to installation in CFM56-2A series engines, inspect uninstalled parts listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-2A SB 72-470, dated November 12, 1999, in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB.

CFM56-2B Series

(14) For CFM56-2B engine nameplate models, with HPT front rotating air seals listed by S/N in paragraph 1.A(1), Effectivity, of CFM56-2B SB 72-611, dated November 12, 1999, inspect in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB, after accumulating 3,000 CSN but before accumulating 6,000 CSN.

Uninstalled Parts

(15) Prior to installation in CFM56-2B series engines, inspect uninstalled parts listed by S/N in paragraph 1.A(1), Effectivity, of CFMI CFM56-2B SB 72-611, dated November 12, 1999, in accordance with the procedures described in Paragraph 2, Accomplishment Instructions, of that SB.

Replace Cracked Parts

(16) Prior to further flight, replace cracked HPT front rotating air seals with serviceable parts.

Definition

(b) For the purpose of this AD, an engine shop visit is defined as the next time, after the effective date of this AD, an engine is in the shop for the purpose of maintenance or inspection.

Alternative Methods of Compliance

(c) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Engine Certification Office (ECO). Operators shall submit their request through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, ECO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this airworthiness directive, if any, may be obtained from the ECO.

Incorporation by Reference

(d) The inspections shall be done in accordance with the following CFMI SB's: CFMI CFM56-3/3B/3C SB 72-922, dated November 12, 1999; CFMI CFM56-2 SB 72-869, dated November 12, 1999; CFM56-2A SB 72-470, dated November 12, 1999, and CFM56-2B SB 72-611, dated November 12, 1999. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from CFM International, Technical Publications Department, 1 Neumann Way, Cincinnati, OH 45215; telephone (513) 552-2800, fax (513) 552-2816. Copies may be inspected at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

Ferry Flights

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the aircraft to a location where the inspection requirements of this AD can be accomplished.

(f) This amendment becomes effective on May 2, 2000.

Issued in Burlington, Massachusetts, on March 7, 2000.

David A. Downey,

Assistant Manager, Engine and Propeller Directorate, Aircraft Certification Service.

[FR Doc. 00-6552 Filed 3-27-00; 8:45 am]

BILLING CODE 4910-13-U

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99-NM-256-AD; Amendment 39-11587; AD 2000-04-05]

RIN 2120-AA64

Airworthiness Directives; Israel Aircraft Industries, Ltd., Model Astra SPX Series Airplanes; Correction

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule; correction.

SUMMARY: This document adds a line of text that was inadvertently omitted from the applicability of airworthiness directive (AD) 2000-04-05 that applies to certain Israel Aircraft Industries, Ltd., Model Astra SPX series airplanes which was published on February 23, 2000 (65 FR 8848). That AD currently requires a one-time inspection to measure the countersink angle of the bolt holes in the lower scissors fitting of the horizontal stabilizer, and corrective actions, if necessary. This document corrects the applicability to include the serial numbers for Model Astra SPX series airplanes. This correction is necessary to ensure that the appropriate operators accomplish the requirements of the AD.

DATES: Effective March 29, 2000.

The incorporation by reference of certain publications listed in the regulations was approved previously by the Director of the Federal Register as of March 29, 2000 (65 FR 8848, February 23, 2000).

FOR FURTHER INFORMATION CONTACT:

Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Airworthiness Directive (AD) 2000-04-05, amendment 39-11587, applicable to certain Israel Aircraft Industries, Ltd., Model Astra SPX series airplanes, was published in the **Federal Register** on February 23, 2000 (65 FR 8848). That AD requires a one-time inspection to measure the countersink angle of the bolt holes in the lower scissors fitting of the horizontal stabilizer, and corrective actions, if necessary.

As published, the applicability of AD 2000-04-05 inadvertently omitted "serial numbers 085 through 112 inclusive" for Israel Aircraft Industries, Ltd., Model Astra SPX series airplanes.

Since no other part of the regulatory information has been changed, the final rule is not being republished.

The effective date of this AD remains March 29, 2000.

§ 39.13 [Corrected]

1. On page 8848, in the third column, the applicability paragraph that precedes Note 1 of AD 2000-04-05 is corrected to read as follows:

* * * * *

Applicability: Model Astra SPX series airplanes, serial numbers 085 through 112 inclusive, certificated in any category.

* * * * *