

directors meetings required under paragraph (b)(1).

3. Amend § 932.17 by:

- a. Revising paragraphs (a) through (c); and
- b. Adding paragraph (f), to read as follows:

§ 932.17 Compensation and expenses of Bank directors.

(a) *Definition.* As used in this section, compensation means any payment of money or provision of any other thing of value (or the accrual of a right to receive money or a thing of value in a subsequent year) in consideration of a director's performance of official duties for the Bank, including, without limitation, daily meeting fees, incentive payments and fringe benefits.

(b) *Annual compensation policy.* Beginning in 2000 and annually thereafter, each Bank's board of directors shall adopt by resolution a written policy to provide for the payment to Bank directors of reasonable compensation for the performance of their duties as members of the Bank's board of directors, subject to the requirements set forth in paragraph (c) of this section. At a minimum, such policy shall address the activities or functions for which attendance is necessary and appropriate and may be compensated, and shall explain and justify the methodology for determining the amount of compensation to be paid to directors.

(c) *Policy requirements.* Payment to directors under each Bank's policy on director compensation may be based upon factors that the Bank determines to be appropriate, but each Bank's policy shall conform to the following requirements:

(1) *Statutory limits on annual compensation.* Pursuant to section 7(i) of the Act, as amended, for 2000, the following limits on compensation shall apply: for a Chairperson—\$25,000; for a Vice Chairperson—\$20,000; for any other member of the Bank's board of directors—\$15,000. Beginning in 2001 and for subsequent years, these limits on annual compensation shall be adjusted annually by the Finance Board to reflect any percentage increase in the preceding year's Consumer Price Index (CPI) for all urban consumers, as published by the Department of Labor. Each year, as soon as practicable after the publication of the previous year's CPI, the Finance Board shall publish notice by **Federal Register**, distribution of a memorandum, or otherwise, of the CPI-adjusted limits on annual compensation.

(2) *Compensation permitted only for performance of official Bank business.*

The total compensation received by each director in a year shall reflect the amount of time spent on official Bank business, such that greater or lesser attendance at board and committee meetings during a given year will be reflected in the compensation received by the director for that year. A Bank shall not pay fees to a director, such as retainer fees, that do not reflect the director's performance of official Bank business.

* * * * *

(f) *Approval.* Payments made to directors in compliance with the limits on annual directors' compensation and the standards set forth in this section are deemed to be approved by the Finance Board for purposes of section 7(i) of the Act, as amended.

4. Remove §§ 932.18 and 932.19, and reserve subpart C.

PART 934—OPERATIONS OF THE BANKS

5. The authority citation for part 934 continues to read as follows:

Authority: 12 U.S.C. 1422a, 1422b, 1431(g), 1432(a), and 1442.

- 6. Amend § 934.7 by:
 - a. Removing the words "and reporting requirements" from the heading;
 - b. Removing paragraphs (a)(2), (b), (c), (d) and (e); and
 - c. Redesignating paragraphs (a)(1), (3), (4) and (5) as paragraphs (a), (b), (c) and (d), respectively.
- 7. Revise § 934.16 to read as follows:

§ 934.16 Bank bylaws.

A Bank's board of directors shall have in effect at all times bylaws governing the manner in which the Bank administers its affairs, and such bylaws shall be consistent with applicable laws and regulations as administered by the Finance Board.

8. Revise § 934.17 to read as follows:

§ 934.17 Bank dividends.

A Bank's board of directors may declare and pay a dividend only from previously retained earnings or current net earnings, and only if such payment will not result in a projected impairment of the par value of the capital stock of the Bank. Dividends on such capital stock shall be computed without preference.

PART 935—ADVANCES

9. The authority citation for part 935 continues to read as follows:

Authority: 12 U.S.C. 1422a(a)(3), 1422b(a)(1), 1426, 1429, 1430, 1430b and 1431.

§ 935.4 [Amended]

- 10. Amend § 935.4 by:
 - a. Removing paragraph designation (d)(1); and
 - b. Removing paragraph (d)(2).

§ 935.5 [Amended]

- 11. Amend § 935.5 by:
 - a. Removing paragraph (a)(2); and
 - b. Redesignating paragraph (a)(3) as paragraph (a)(2).

§ 935.16 [Removed]

12. Remove § 935.16.

Dated: December 14, 1999.

By the Board of Directors of the Federal Housing Finance Board.

Bruce A. Morrison,
Chairman.

[FR Doc. 99-33069 Filed 12-20-99; 8:45 am]

BILLING CODE 6725-01-P

DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 96-NM-194-AD; Amendment 39-11467; AD 99-26-08]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A310 and A300-600 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Airbus Model A310 and A300-600 series airplanes, that requires replacement of the rudder trim switch in the flight compartment with a new switch having a longer shaft; modification of wiring in panel 408VU; and replacement of the rudder trim control knob with an improved new knob. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to prevent inadvertent and uncommanded rudder trim activation, which could result in yaw and roll excursions and consequent reduced controllability of the airplane.

DATES: Effective January 25, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of January 25, 2000.

ADDRESSES: The service information referenced in this AD may be obtained

from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Norman B. Martenson, Manager, International Branch, ANM-116, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Airbus Model A310 and A300-600 series airplanes was published as a supplemental notice of proposed rulemaking (NPRM) in the **Federal Register** on February 12, 1998 (63 FR 7076). That action proposed to require replacement of the rudder trim switch in the flight compartment with a new switch having a longer shaft; modification of wiring in panel 408VU; and replacement of the rudder trim control knob with an improved new knob.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the supplemental NPRM or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

The FAA estimates that 90 airplanes of U.S. registry will be affected by this AD.

Replacement of the rudder trim switch and modification of the wiring will take approximately 7 work hours per airplane to accomplish, at an average labor rate of \$60 per work hour. Required parts will be provided by the manufacturer at no cost to the operators. Based on these figures, the cost impact of this action on U.S. operators is estimated to be \$37,800, or \$420 per airplane.

Replacement of the rudder trim control knob will take approximately 1 work hour per airplane to accomplish, at an average labor rate of \$60 per work hour. Required parts will be provided

by the manufacturer at no cost to the operators. Based on these figures, the cost impact of this action on U.S. operators is estimated to be \$5,400, or \$60 per airplane.

The cost impact figures discussed above are based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government.

Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption **ADDRESSES**.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

99-26-08 AIRBUS INDUSTRIE: Amendment 39-11467. Docket 96-NM-194-AD.

Applicability: Model A310 and A300-600 series airplanes, certificated in any category; except those on which Airbus Modification 11874 [reference Airbus Service Bulletin A310-27-2087 (for Model A300 series airplanes) or A300-27-6042 (for Model A300-600 series airplanes), both dated October 2, 1998] has been accomplished.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (d) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent inadvertent and uncommanded rudder trim activation, which could result in yaw and roll excursions and consequent reduced controllability of the airplane, accomplish the following:

Corrective Actions

(a) Within 90 days after the effective date of this AD, replace the rudder trim switch, part number (P/N) 097-023-00, in the flight compartment, with a new switch, P/N 097-023-01; and modify the wiring in panel 408VU; in accordance with Airbus Service Bulletin A310-27-2084, Revision 01 (for Model A310 series airplanes); or A300-27-6037, Revision 01 (for Model A300-600 series airplanes), both dated September 29, 1998; as applicable.

Note 2: Accomplishment of the actions required by paragraph (a) of this AD in accordance with Airbus Service Bulletin A310-27-2084 (for Model A310 series airplanes); or A300-27-6037 (for Model A300-600 series airplanes), both dated February 12, 1997; as applicable, is acceptable for compliance with that paragraph.

(b) Within 10 months after the effective date of this AD, replace the rudder trim control knob on the rudder trim switch with an improved new knob in accordance with Airbus Service Bulletin A310-27-2087, Revision 01 (for Model A310 series airplanes); or A300-27-6042, Revision 01 (for Model A300-600 series airplanes), both dated February 17, 1999; as applicable.

Note 3: Accomplishment of the actions required by paragraph (b) of this AD in accordance with Airbus Service Bulletin A310-27-2087 (for Model A310 series airplanes); or A300-27-6042 (for Model

A300–600 series airplanes), both dated October 2, 1998; as applicable, is acceptable for compliance with that paragraph.

Spares

(c) As of the effective date of this AD, no person shall install in the flight compartment of any airplane a rudder trim switch having P/N 097–023–00.

Alternative Methods of Compliance

(d) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM–116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM–116.

Note 4: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM–116.

(e) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(f) The actions shall be done in accordance with Airbus Service Bulletin A310–27–2084, Revision 01, dated September 29, 1998; Airbus Service Bulletin A300–27–6037, Revision 01, dated September 29, 1998; Airbus Service Bulletin A310–27–2087, Revision 01, dated February 17, 1999; or Airbus Service Bulletin A300–27–6042, Revision 01, dated February 17, 1999; as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 5: The subject of this AD is addressed in French airworthiness directives 97–111–219(B), dated May 7, 1997, and 1999–012–275(B), dated January 13, 1999.

(g) This amendment becomes effective on January 25, 2000.

Issued in Renton, Washington, on December 9, 1999.

D.L. Riggins,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 99–32508 Filed 12–20–99; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 99–NE–32–AD; Amendment 39–11465; AD 99–26–06]

RIN 2120–AA64

Airworthiness Directives; Pratt & Whitney JT8D–200 Series Turbofan Engines

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to Pratt & Whitney JT8D–200 series turbofan engines, that requires initial and repetitive fluorescent magnetic particle inspections or fluorescent penetrant inspections of the combustion chamber outer case (CCOC) for cracks, and, if necessary, replacement with serviceable parts. Also, this AD requires a one-time boss material verification, and, if necessary, replacement with serviceable parts. Finally, this AD requires replacement of CCOCs with welded-on bosses with improved, one-piece CCOCs. Installation of the one-piece CCOC constitutes terminating action to the inspection requirements of this AD. This amendment is prompted by a report of an uncontained engine failure caused by fatigue cracks originating at the weld joining the drain boss to the CCOC. The actions specified by this AD are intended to prevent CCOC cracks, which could result in an uncontained engine failure and damage to the airplane.

DATES: Effective February 22, 2000.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of February 22, 2000.

ADDRESSES: The service information referenced in this AD may be obtained from Pratt & Whitney, 400 Main St., East Hartford, CT 06108; telephone (860) 565–8770, fax (860) 565–4503. This information may be examined at the Federal Aviation Administration (FAA), New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW, suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT: Christopher Spinney, Aerospace Engineer, Engine Certification Office, FAA, Engine and Propeller Directorate,

12 New England Executive Park, Burlington, MA 01803–5299; telephone (781) 238–7175, fax (781) 238–7199.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to Pratt & Whitney (PW) JT8D–209, –217, –217A, –217C, and –219 series turbofan engines was published in the **Federal Register** on September 23, 1999 (64 FR 51483). That action proposed to require initial and repetitive fluorescent magnetic particle inspections or fluorescent penetrant inspections of the combustion chamber outer case (CCOC) for cracks, and, if necessary, replacement with serviceable parts. Also, that AD proposed to require a one-time boss material verification, and, if necessary, replacement with serviceable parts. Finally, that AD proposed to require replacement of CCOCs with welded-on bosses with improved, one-piece CCOCs. Installation of the one-piece CCOC would constitute terminating action to the inspection requirements of the AD. That action was prompted by a report of an uncontained engine failure caused by fatigue cracks originating at the weld joining the drain boss to the CCOC. That condition, if not corrected, could result in CCOC cracks, which could result in an uncontained engine failure and damage to the airplane.

Comments Received

Interested persons have been afforded an opportunity to participate in the making of this amendment. Due consideration has been given to the comments received.

Understated Financial Impact

One commenter states that the Federal Aviation Administration (FAA) has understated the financial impact of the AD by not including the ancillary costs of removing a cracked CCOC. The FAA does not concur. The indirect costs associated with this AD are not directly related to this rule, and, therefore, are not addressed in the economic analysis for this rule. A full cost analysis for each AD, including such indirect costs, is not necessary since the FAA has already performed a cost benefit analysis when adopting the airworthiness requirements to which these engines were originally certificated. A finding that an AD is warranted means that the original design no longer achieves the level of safety specified by those airworthiness requirements, and that other required actions are necessary, such as inspections of existing CCOCs and replacement with a one-piece CCOC. Because the original level of safety was