

(3) If the crack is greater than 0.118 inches (5.0 mm) in length, inspect, ream, cold work, and replace prior to further flight.

Optional Terminating Modification

(c) Accomplishment of the high frequency eddy current inspection for fatigue cracking of the holes in the wing structure; reaming and cold working of those holes; and replacement of all aluminum wing spoiler number 3 actuator attachment fittings with new steel fittings; in accordance with Airbus Industrie Service Bulletin A310-57-2079, Revision 01, dated January 11, 1999; constitutes terminating action for the repetitive inspection requirements of paragraph (a) of this AD.

Note 3: Replacement of aluminum attachment fittings of the number 3 wing spoilers with steel fittings accomplished prior to the effective date of this AD in accordance with the original issue of Airbus Industrie Service Bulletin A310-57-2079, dated July 21, 1998, is considered acceptable for compliance with the applicable fitting replacement specified in paragraphs (b) and (c) of this AD.

Wing Repair

(d) If any crack is found in the wing structure during any inspection required by paragraph (b) or specified in paragraph (c) of this AD, prior to further flight, repair in accordance with a method approved by either the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate; or the Direction Générale de l'Aviation Civile (DGAC) (or its delegated agent). For a repair method to be approved by the Manager, International Branch, ANM-116, as required by this paragraph, the Manager's approval letter must specifically reference this AD.

Note 4: For paragraph (d) of this AD, the wing spoiler number 3 actuator attachment fittings are not considered part of the wing structure.

Alternative Methods of Compliance

(e) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 5: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(f) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(g) Except as provided by paragraph (d) of this AD, the actions shall be done in

accordance with Airbus Industrie Service Bulletin A310-57-2078, Revision 01, dated January 11, 1999; or Airbus Industrie Service Bulletin A310-57-2079, Revision 01, dated January 11, 1999; as applicable. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 6: The subject of this AD is addressed in French airworthiness directive 98-483-271(B) R1, dated June 2, 1999.

(h) This amendment becomes effective on October 27, 1999.

Issued in Renton, Washington, on September 10, 1999.

D.L. Riffin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 99-24199 Filed 9-21-99; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-NM-384-AD; Amendment 39-11324; AD 99-19-37]

RIN 2120-AA64

Airworthiness Directives; Bombardier Model DHC-8-100 and -300 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Bombardier Model DHC-8-100 and -300 series airplanes, that requires replacement of the main landing gear (MLG) uplock actuator on both the left and right MLG with a new redesigned uplock assembly. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to prevent failure of the MLG to extend when a "gear down" selection is made.

DATES: Effective October 27, 1999.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of October 27, 1999.

ADDRESSES: The service information referenced in this AD may be obtained

from Bombardier, Inc., Bombardier Regional Aircraft Division, Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Engine and Propeller Directorate, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

FOR FURTHER INFORMATION CONTACT:

Paolo Farina, Aerospace Engineer, Systems and Flight Test Branch, ANE-172, FAA, Engine and Propeller Directorate, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York 11581; telephone (516) 256-7530; fax (516) 568-2716.

SUPPLEMENTARY INFORMATION: A proposal to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) to include an airworthiness directive (AD) that is applicable to certain Bombardier Model DHC-8-100 and -300 series airplanes was published in the **Federal Register** on July 20, 1999 (64 FR 38850). That action proposed to require replacement of the main landing gear (MLG) uplock actuator on both the left and right MLG with a new redesigned uplock assembly.

Comments

Interested persons have been afforded an opportunity to participate in the making of this amendment. No comments were submitted in response to the proposal or the FAA's determination of the cost to the public.

Conclusion

The FAA has determined that air safety and the public interest require the adoption of the rule as proposed.

Cost Impact

The FAA estimates that 148 airplanes of U.S. registry will be affected by this AD, that it will take approximately 6 work hours per airplane to accomplish the required replacement, and that the average labor rate is \$60 per work hour. Required parts will cost between \$4,030 and \$5,016 per airplane. Based on these figures, the cost impact of the required AD on U.S. operators is estimated to be between \$649,720 and \$795,648, or between \$4,390 and \$5,376 per airplane.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the requirements of this AD action, and that no operator would accomplish

those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations adopted herein will not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this final rule does not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this action (1) is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A final evaluation has been prepared for this action and it is contained in the Rules Docket. A copy of it may be obtained from the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Incorporation by reference, Safety.

Adoption of the Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration amends part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

99-19-37 Bombardier, Inc. (Formerly de Havilland, Inc.): Amendment 39-11324. Docket 98-NM-384-AD.

Applicability: Model DHC-8-100 and -300 series airplanes, serial numbers 3 through 339 inclusive, except those on which Modification 8/1828 has been incorporated; certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in

the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent failure of the main landing gear to extend when a "gear down" selection is made, accomplish the following:

(a) Within 12 months after the effective date of this AD: Replace the uplock actuator with a new, improved part in accordance with de Havilland Service Bulletin S.B. 8-32-98, Revision 'C,' dated July 31, 1998.

Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, New York Aircraft Certification Office (ACO), FAA, Engine and Propeller Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, New York ACO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the New York ACO.

Special Flight Permits

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Incorporation by Reference

(d) The replacement shall be done in accordance with de Havilland Service Bulletin S.B. 8-32-98, Revision 'C,' dated July 31, 1998. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Bombardier, Inc., Bombardier Regional Aircraft Division, Garratt Boulevard, Downsview, Ontario M3K 1Y5, Canada. Copies may be inspected at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW., Renton, Washington; or at the FAA, Engine and Propeller Directorate, New York Aircraft Certification Office, 10 Fifth Street, Third Floor, Valley Stream, New York; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

Note 3: The subject of this AD is addressed in Canadian airworthiness directive CF-98-26, dated August 26, 1998.

(e) This amendment becomes effective on October 27, 1999.

Issued in Renton, Washington, on September 10, 1999.

D.L. Riggin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.
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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-NM-366-AD; Amendment 39-11323; AD 99-19-36]

RIN 2120-AA64

Airworthiness Directives; Dornier Model 328-100 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Final rule.

SUMMARY: This amendment adopts a new airworthiness directive (AD), applicable to certain Dornier Model 328-100 series airplanes, that requires a one-time inspection to measure the offset of the de-icing tubing adjacent to the refueling panel on the right-hand wing, and replacement with new improved tubing, if necessary. This amendment is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by this AD are intended to prevent a blockage in the de-icing tubing which could result in a malfunction of the de-icing boot. This malfunction would be unknown to the flight crew, and could lead to reduced controllability of the airplane during flight in icing conditions.

DATES: Effective October 27, 1999.

The incorporation by reference of certain publications listed in the regulations is approved by the Director of the Federal Register as of October 27, 1999.

ADDRESSES: The service information referenced in this AD may be obtained from Fairchild Dornier, Dornier Luftfahrt GmbH, P.O. Box 1103, D-82230 Wessling, Germany. This information may be examined at the Federal Aviation Administration (FAA), Transport Airplane Directorate, Rules Docket, 1601 Lind Avenue, SW., Renton, Washington; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.