

December 1, 1998, to detect cracking around certain fastener holes and adjacent areas of the front spar of the horizontal stabilizer, in accordance with the service bulletin, at the time specified in paragraph (b)(1), (b)(2), or (b)(3) of this AD, as applicable. Thereafter, repeat only the eddy current and x-ray inspections at intervals not to exceed 6,000 flight cycles until the requirements of paragraph (d) of this AD are accomplished.

(1) For airplanes that have accumulated less than 12,000 total flight cycles as of the effective date of this AD: Perform an eddy current and an x-ray inspection prior to the accumulation of 12,000 total flight cycles, or within 2,000 flight cycles after the effective date of this AD, whichever occurs later.

(2) For airplanes that have accumulated 12,000 or more total flight cycles and less than 16,000 total flight cycles as of the effective date of this AD: Accomplish the requirements of paragraphs (b)(2)(i) and (b)(2)(ii) of this AD.

(i) Perform a detailed visual inspection within 800 flight cycles after the effective date of this AD; and

(ii) Perform an eddy current and an x-ray inspection within 2,000 flight cycles after the effective date of this AD.

(3) For airplanes that have accumulated 16,000 or more total flight cycles as of the effective date of this AD: Accomplish the requirements of paragraphs (b)(3)(i) and (b)(3)(ii) of this AD.

(i) Perform a detailed visual inspection within 400 flight cycles after the effective date of this AD; and

(ii) Perform an eddy current and an x-ray inspection within 1,200 flight cycles after the effective date of this AD.

Corrective Actions

(c) If any cracking is detected during any inspection required by paragraph (a) or (b) of this AD, prior to further flight, either repair in accordance with a method approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, or the Luftfartsverket (LFV) (or its delegated agent); or accomplish the requirements of paragraph (d) of this AD.

Note 2: Inspections to detect cracking around certain fastener holes and adjacent areas of the front spar of the horizontal stabilizers that have been accomplished prior to the effective date of this AD in accordance with Saab Service Bulletin 340-55-033, Revision 03, dated January 22, 1998, are considered acceptable for compliance with the applicable action specified by this AD.

Terminating Action

(d) For all airplanes: Except as provided by paragraph (e) of this AD, accomplish cold working of certain fastener holes of the front spar of the horizontal stabilizers, and follow-on actions; and install new fasteners; in accordance with Saab Service Bulletin 340-55-034, dated October 16, 1998; at the time specified in paragraph (d)(1), (d)(2), or (d)(3) of this AD, as applicable. Accomplishment of this action constitutes terminating action for this AD.

(1) For all airplanes that have accumulated less than 26,000 total flight cycles as of the effective date of this AD: Within 10,000 flight cycles after the effective date of this AD.

(2) For all airplanes that have accumulated 26,000 or more total flight cycles and less than 30,000 total flight cycles as of the effective date of this AD: Within 6,000 flight cycles after the effective date of this AD.

(3) For all airplanes that have accumulated 30,000 or more total flight cycles as of the effective date of this AD: Within 3,000 flight cycles after the effective date of this AD.

(e) If any crack is detected during the accomplishment of paragraph (d) of this AD, and if the service bulletin listed in paragraph (d) of this AD specifies to contact the manufacturer for an appropriate corrective action: Prior to further flight, repair in accordance with a method approved by the Manager, International Branch, ANM-116, or the LFV (or its delegated agent).

Alternative Method of Compliance

(f) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 3: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(g) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Note 4: The subject of this AD is addressed in Swedish airworthiness directives 1-110R2, dated December 7, 1998, and 1-133, dated October 20, 1998.

Issued in Renton, Washington, on July 9, 1999.

D.L. Riggins,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

[FR Doc. 99-18100 Filed 7-14-99; 8:45 am]

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DEPARTMENT OF TRANSPORTATION

Federal Aviation Administration

14 CFR Part 39

[Docket No. 98-NM-344-AD]

RIN 2120-AA64

Airworthiness Directives; British Aerospace BAe Model ATP Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness

directive (AD) that is applicable to all British Aerospace BAe Model ATP airplanes. This proposal would require repetitive tests for the serviceability of the nose landing gear compensator; and corrective action, if necessary. This proposal is prompted by issuance of mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by the proposed AD are intended to prevent a nose wheel shimmy, which could lead to the collapse of the nose landing gear during landing.

DATES: Comments must be received by August 16, 1999.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 98-NM-344-AD, 1601 Lind Avenue, SW, Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from AI(R) American Support, Inc., 13850 Mclearen Road, Herndon, Virginia 20171. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW, Renton, Washington.

FOR FURTHER INFORMATION CONTACT: Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW, Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:

Comments Invited

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments, specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report

summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to Docket Number 98-NM-344-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 98-NM-344-AD, 1601 Lind Avenue, SW, Renton, Washington 98055-4056.

Discussion

The Civil Aviation Authority (CAA), which is the airworthiness authority for the United Kingdom, notified the FAA that an unsafe condition may exist on all British Aerospace BAe Model ATP airplanes. The CAA advises that a nose wheel shimmy resulted in the collapse of the nose landing gear during landing. The steering compensator was found to be unserviceable due to a high leakage rate of hydraulic fluid through the unit. The existing design of the nose landing gear steering system may not adequately prevent such leakage. This condition, if not corrected, could lead to the collapse of the nose landing gear.

Explanation of Relevant Service Information

British Aerospace has issued Alert Service Bulletin ATP-A32-94, dated October 3, 1998, which describes procedures for repetitive tests for the serviceability of the nose landing gear compensator, and replacement of the compensator with a new or serviceable part, if necessary. Accomplishment of the actions specified in the service bulletin is intended to adequately address the identified unsafe condition. The CAA classified this service bulletin as mandatory and issued British airworthiness directive 016-10-98, in order to assure the continued airworthiness of these airplanes in the United Kingdom.

FAA's Conclusions

This airplane model is manufactured in the United Kingdom and is type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral

airworthiness agreement, the CAA has kept the FAA informed of the situation described above. The FAA has examined the findings of the CAA, reviewed all available information, and determined that AD action is necessary for products of this type design that are certificated for operation in the United States.

Explanation of Requirements of Proposed Rule

Since an unsafe condition has been identified that is likely to exist or develop on other airplanes of the same type design registered in the United States, the proposed AD would require accomplishment of the actions specified in the service bulletin described previously.

Cost Impact

The FAA estimates that 10 airplanes of U.S. registry would be affected by this proposed AD, that it would take approximately 2 work hours per airplane to accomplish the proposed test, and that the average labor rate is \$60 per work hour. Required parts would cost approximately \$50 per airplane. Based on these figures, the cost impact of the proposed AD on U.S. operators is estimated to be \$1,700, or \$170 per airplane, per test.

The cost impact figure discussed above is based on assumptions that no operator has yet accomplished any of the proposed requirements of this AD action, and that no operator would accomplish those actions in the future if this AD were not adopted.

Regulatory Impact

The regulations proposed herein would not have substantial direct effects on the States, on the relationship between the national government and the States, or on the distribution of power and responsibilities among the various levels of government. Therefore, in accordance with Executive Order 12612, it is determined that this proposal would not have sufficient federalism implications to warrant the preparation of a Federalism Assessment.

For the reasons discussed above, I certify that this proposed regulation (1) Is not a "significant regulatory action" under Executive Order 12866; (2) is not a "significant rule" under the DOT Regulatory Policies and Procedures (44 FR 11034, February 26, 1979); and (3) if promulgated, will not have a significant economic impact, positive or negative, on a substantial number of small entities under the criteria of the Regulatory Flexibility Act. A copy of the draft regulatory evaluation prepared for this action is contained in the Rules Docket.

A copy of it may be obtained by contacting the Rules Docket at the location provided under the caption ADDRESSES.

List of Subjects in 14 CFR Part 39

Air transportation, Aircraft, Aviation safety, Safety.

The Proposed Amendment

Accordingly, pursuant to the authority delegated to me by the Administrator, the Federal Aviation Administration proposes to amend part 39 of the Federal Aviation Regulations (14 CFR part 39) as follows:

PART 39—AIRWORTHINESS DIRECTIVES

1. The authority citation for part 39 continues to read as follows:

Authority: 49 U.S.C. 106(g), 40113, 44701.

§ 39.13 [Amended]

2. Section 39.13 is amended by adding the following new airworthiness directive:

BRITISH AEROSPACE REGIONAL

AIRCRAFT [Formerly Jetstream Aircraft Limited; British Aerospace (Commercial Aircraft) Limited]: Docket 98-NM-344-AD.

Applicability: All BAe Model ATP airplanes, certificated in any category.

Note 1: This AD applies to each airplane identified in the preceding applicability provision, regardless of whether it has been modified, altered, or repaired in the area subject to the requirements of this AD. For airplanes that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and, if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To prevent a nose wheel shimmy, which could lead to the collapse of the nose landing gear during landing, accomplish the following:

Serviceability Test

(a) Within 250 flight cycles after the effective date of this AD, perform a test for the serviceability of the nose landing gear compensator in accordance with British Aerospace Alert Service Bulletin ATP-A32-94, dated October 3, 1998. Thereafter, repeat the test at intervals not to exceed 4,000 flight cycles. If the compensator does not pass the serviceability test, within 50 flight cycles after the accomplishment of the test, replace the compensator with a new or serviceable compensator in accordance with the service bulletin.

Alternative Methods of Compliance

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate. Operators shall submit their requests through an appropriate FAA Principal Maintenance Inspector, who may add comments and then send it to the Manager, International Branch, ANM-116.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the International Branch, ANM-116.

Special Flight Permits

(c) Special flight permits may be issued in accordance with sections 21.197 and 21.199 of the Federal Aviation Regulations (14 CFR 21.197 and 21.199) to operate the airplane to a location where the requirements of this AD can be accomplished.

Note 3: The subject of this AD is addressed in British airworthiness directive 016-10-98.

Issued in Renton, Washington, on July 9, 1999.

D.L. Riggin,

Acting Manager, Transport Airplane Directorate, Aircraft Certification Service.

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DEPARTMENT OF TRANSPORTATION**Federal Aviation Administration****14 CFR Part 39**

[Docket No. 99-NM-110-AD]

RIN 2120-AA64

Airworthiness Directives; Airbus Model A310 and A300-600 Series Airplanes

AGENCY: Federal Aviation Administration, DOT.

ACTION: Notice of proposed rulemaking (NPRM).

SUMMARY: This document proposes the adoption of a new airworthiness directive (AD) that is applicable to certain Airbus Model A310 and A300-600 series airplanes. This proposal would require a detailed visual inspection to detect damage to the terminal lugs on the 12XC and 15XE connectors and the mounting lugs on the 15XE connector; and repair or replacement of the terminal lugs or the 15XE connector with new parts, if necessary. This proposal is prompted by the issuance of a mandatory continuing airworthiness information by a foreign civil airworthiness authority. The actions specified by the proposed AD are intended to detect and correct broken terminal and mounting lugs on

the 15XE and 12XC connectors in the 101VU panel in the avionics compartment, which could result in loss of electrical power from the standby generator.

DATES: Comments must be received by August 16, 1999.

ADDRESSES: Submit comments in triplicate to the Federal Aviation Administration (FAA), Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 99-NM-110-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056. Comments may be inspected at this location between 9 a.m. and 3 p.m., Monday through Friday, except Federal holidays.

The service information referenced in the proposed rule may be obtained from Airbus Industrie, 1 Rond Point Maurice Bellonte, 31707 Blagnac Cedex, France. This information may be examined at the FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW, Renton, Washington.

FOR FURTHER INFORMATION CONTACT: Norman B. Martenson, Manager, International Branch, ANM-116, FAA, Transport Airplane Directorate, 1601 Lind Avenue, SW, Renton, Washington 98055-4056; telephone (425) 227-2110; fax (425) 227-1149.

SUPPLEMENTARY INFORMATION:**Comments Invited**

Interested persons are invited to participate in the making of the proposed rule by submitting such written data, views, or arguments as they may desire. Communications shall identify the Rules Docket number and be submitted in triplicate to the address specified above. All communications received on or before the closing date for comments specified above, will be considered before taking action on the proposed rule. The proposals contained in this notice may be changed in light of the comments received.

Comments are specifically invited on the overall regulatory, economic, environmental, and energy aspects of the proposed rule. All comments submitted will be available, both before and after the closing date for comments, in the Rules Docket for examination by interested persons. A report summarizing each FAA-public contact concerned with the substance of this proposal will be filed in the Rules Docket.

Commenters wishing the FAA to acknowledge receipt of their comments submitted in response to this notice must submit a self-addressed, stamped postcard on which the following statement is made: "Comments to

Docket Number 99-NM-110-AD." The postcard will be date stamped and returned to the commenter.

Availability of NPRMs

Any person may obtain a copy of this NPRM by submitting a request to the FAA, Transport Airplane Directorate, ANM-114, Attention: Rules Docket No. 99-NM-110-AD, 1601 Lind Avenue, SW., Renton, Washington 98055-4056.

Discussion

The Direction Generale de L'Aviation Civile (DGAC), which is the airworthiness authority for France, notified the FAA that an unsafe condition may exist on certain Airbus Model A310 and A300-600 series airplanes. The DGAC advises that operators have reported cases of broken terminal lugs on the three-phase wiring to the 12XC and 15XE connectors located on the essential part of the 101VU panel in the avionics compartment on airplanes equipped with a standby generator. The DGAC also advises that the mounting lugs on the 15XE connector were also found broken. Preliminary indications are that the mounting configuration of connector 15XE is transmitting vibration to the terminal lugs of both connectors and to the mounting lugs of 15XE. Such failures, if not detected and corrected, could result in loss of electrical power from the standby generator.

Explanation of Relevant Service Information

Airbus has issued an All Operators Telex (AOT) 24-09, Revision 01, dated August 13, 1998, which describes procedures for inspecting the terminal lugs on the 12XC and 15XE connectors and the mounting lugs on the 15XE connector for damage; and repair or replacement of the terminal lugs or the 15XE connector with new parts, if necessary.

The DGAC classified Airbus AOT 24-09 as mandatory and issued French airworthiness directive 1999-077-278(B), dated February 24, 1999, in order to assure the continued airworthiness of these airplanes in France.

FAA's Conclusions

These airplane models are manufactured in France and are type certificated for operation in the United States under the provisions of section 21.29 of the Federal Aviation Regulations (14 CFR 21.29) and the applicable bilateral airworthiness agreement. Pursuant to this bilateral airworthiness agreement, the DGAC has kept the FAA informed of the situation